

CLAIM AMENDMENTS

Please cancel claims 15, 25 and 28.

Please add new claims 29-38.

1. - 15. (Canceled)

16. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 29 45, wherein the cross sectional configurations of the said inlets and said outlet and the geometry of the internal space are designed such that the gas stream circulating in the internal space has a mass flow rate greater ~~is less~~ than twice a mass flow rate of the fresh air introduced into the inlet.

17. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 29 45, wherein the cross sectional configurations of said inlets ~~the inlet~~ and said outlet and the geometry of the internal space are designed such that the gas stream circulating in the internal space has a mass flow rate ~~is larger~~ greater than twice a mass flow rate of the fresh air introduced into the inlet.

18. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 29 45, wherein said inlets include ~~the inlet includes~~ a plurality of air nozzles arranged in said by side relation next to each other in a row.

19. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 18, wherein each air nozzle has a portion extending beyond the wall.

20. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 18, wherein the air nozzles have a ~~corresponding~~ common orientation.
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21. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 18, wherein the combustion chamber has a cylindrical configuration and the air nozzles are arranged on a circle that is arranged concentric to combustion chamber.

22. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 29 45, wherein the combustion chamber is designed as a circular ring.

23. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 29 ~~15~~, wherein said inlets ~~the inlet~~ and said the outlet are arranged and the geometry of the internal space is configured such that the circulating gas stream flow encompasses the entire internal space.

24. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 29 ~~15~~, wherein ~~the~~ a circulating gas stream flow generated in said internal space has only a single turbulence center.

25. (Canceled)

26. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 15, wherein the combustion chamber includes a preheating device.

27. (Currently Amended) The gas turbine compressor ~~combustion chamber~~ according to claim 15, wherein a guide device is arranged in the internal space that divides the internal space into a mixing and reaction channel and a backflow channel.

28. (Canceled)

29. (New) A gas turbine comprising:

a compressor;

a turbine;

a combustion chamber having a wall that encloses and defines an internal reaction space having a longitudinal axis, said combustion chamber having inlets for directing air and/or fuel into said internal space in substantially parallel relation to said longitudinal axis, said combustion chamber having an outlet coaxially located on said longitudinal axis and connected to said turbine for discharging exhaust gases to said turbine,

said internal space being configured such that a relatively large circulating gas flow stream can be formed in the internal space so as to maintain a flameless oxidation process, and

said inlets each being disposed a distance from said longitudinal axis greater than the radius of said discharge orifice.

30. (New) The gas turbine of claim 29 in which said inlets direct air and fuel into said internal reaction chamber in the same direction.

31. (New) The gas turbine of claim 29 in which said inlets include a plurality of air inlets disposed in a circular array having a diameter greater than the diameter of said outlet.

32. (New) The gas turbine of claim 29 in which said inlets and outlet communicate with different axial ends of said internal reaction space.

33. (New) The gas turbine of claim 29 in which said inlets and outlet communicate with a common axial end of said internal reaction space.

34. (New) The gas turbine of claim 29 including a fuel feed device for directing fuel into and through said inlets.

35. (New) A gas turbine comprising:

a compressor;

a turbine;

a combustion chamber having a wall that encloses and defines an internal reaction space having a longitudinal axis, said combustion chamber having air inlets connected to said compressor for directing air into said internal space, said combustion chamber having an outlet coaxially located an said longitudinal axis connected to said turbine for discharging exhaust gases to said turbine

a fuel feed device for directing fuel into said internal space in a predetermined direction,

said air inlets and said fuel feed device being operative for directing air and fuel into said internal reaction space in substantially parallel relation to said longitudinal axis,

said internal space being configured such that a relatively large circulating gas flow stream can be formed in the internal space so as to maintain a flameless oxidation process, and

said air inlets defining a diameter greater than the diameter of said exhaust gas outlet.

36. (New) The gas turbine of claim 35 in which said air inlets and fuel feed device direct air and fuel are introduced into said internal reaction chamber in substantially the same direction.

37. (New) The gas turbine of claim 35 in which said air inlets are disposed in a circular array having a diameter greater than the diameter of said outlet.

38. (New) The gas turbine of claim 35 in which said inlets and outlet communicate with different with different axial ends of said reaction space.